

Future Exploration Missions of ISRO

Dr M Annadurai, Director, ISAC, ISRO Email id – <u>madurai@isac.gov.in</u>

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Missions in Operation

Mars Orbiter Mission

- India's first interplanetary mission
- Orbiting Mars since September 2014
- Completed 32 months in Orbit
- First year Science data released
- 200 GB data downloaded by 1100 users

ASTROSAT

- First Indian ASTROnomy SATellite
- Simultaneous, Multi wavelength Observations
- Launched in September 2015
- Completed 20 months in Orbit
- Payloads from Indian Academia, UK & Canada
- Open to Scientists/Researchers

http://astrosat.iucaa.in/

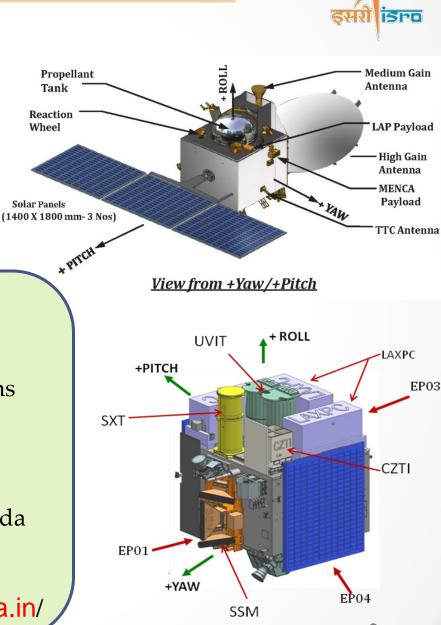
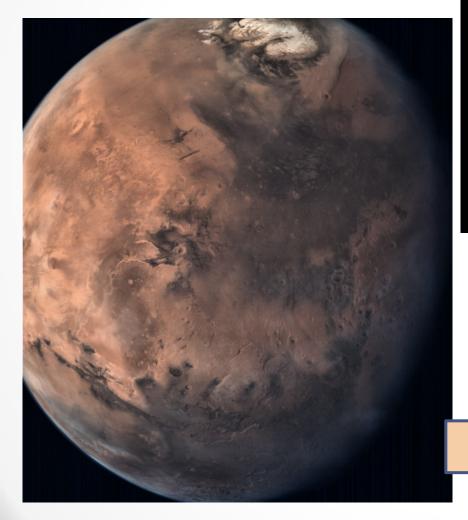
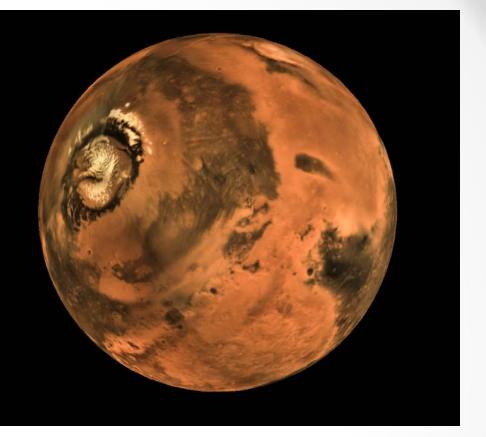


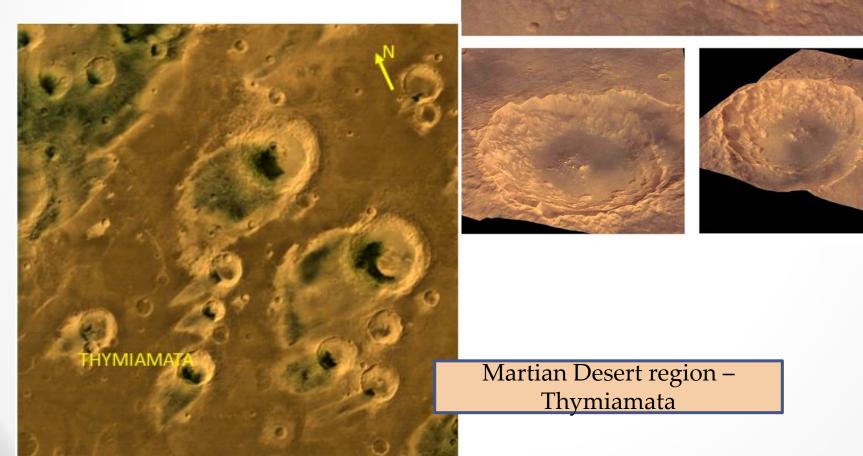
Image from Mars Global Mosaic





Mars Full disc image

Impact crater located SW of Huygens crater



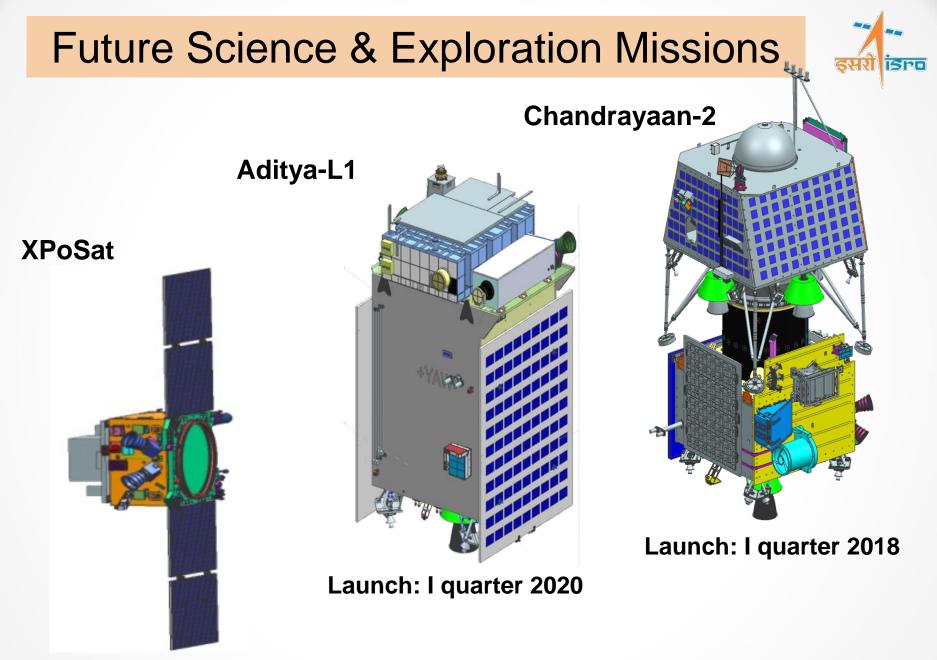
http://www.issdc.gov.in/docs/mr1/Mars-atlas-MOM.pdf

Mars Orbiter Mission (MOM)

Mars Atlas



Space Applications Centre Indian Space Research Organisation

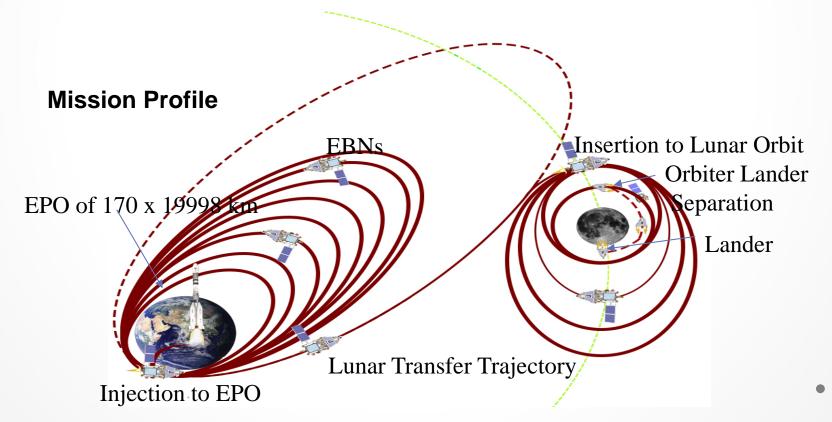


Launch: II quarter 2021

Chandrayaan – 2 Objectives



- Expand technologies from Chandrayaan-1 & demonstrate newer technologies for future planetary missions.
- Deploy a Lunar Lander-Rover capable of soft landing on a specified lunar site and deploy a Rover to carry out in-situ analysis of chemicals.
- Carry payloads in the Orbiter craft to enhance the scientific objectives of Chandrayaan-1 with improved resolution.



	Chandra	yaan – 2 Configuration	TRÌ ISPO	
	ROVER	ORBITER CRAFT		
	Launch vehicle	GSLV MK-II		
	Mass	Composite - 3320 kg		
	Orbiter	100 km Lunar orbit, Mission life - 1 year Orbiter to communicate with IDSN & Lander		
•	Lander	Soft Landing on Moon Mission life of one lunar day (<u>~</u> 14 Earth Days) Communication with IDSN /Orbiter/Rover	•	

	Payloads	Science Objectives	
	TMC-2	Prepares a detailed three dimensional map of the lunar surface.	
	CLASS	Maps the abundance of major rock forming elements Mg, Al, Si, Ca, Ti and Fe on the lunar surface.	
	XSM	Observe the X-rays emitted from the sun and Sun corona and supports CLASS.	
	OHRC	Provides high resolution image of Landing site	
	IIRS	Identifies the minerals & signatures of hydroxyl (OH) and water (H_2O) molecules in polar regions.	
	SAR	Maps lunar craters and other features especially in the polar regions.	
	CHACE -2	Neutral Mass Spectrometer which will carry out a detailed study of the lunar exosphere.	
• 🌃 🦥	RAMBHA	To measure total electron content (TEC) •	
Orbiter		Á.	L
			R (L
		Lander	C



Chandrayaan – 2 Payloads

	Á.	Lander Payloads	Science Objectives
		RAMBHA (LP+DFRS)	Measure TEC of the lunar ionosphere and its morphology. Measure near surface plasma density and its changes with time
Lander		ChaSTE	Measurement of thermal properties of lunar regolith near polar region.
		ILSA	Measure lunar seismicity around the landing site and structure of the lunar crust & mantle.
		Rover Payloads	Science Objectives
D		LIBS	Derive the chemical composition and infers - mineralogical composition on lunar surface
Rover		AÞXS	Determine the elemental composition of lunar rocks and soil

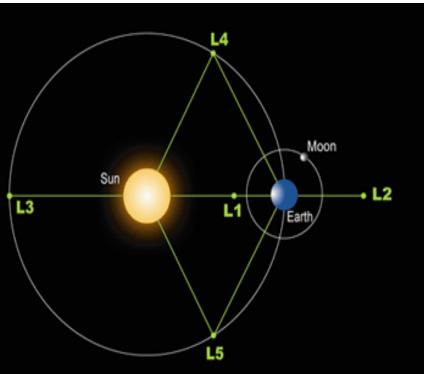
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Aditya-L1 Mission Objectives

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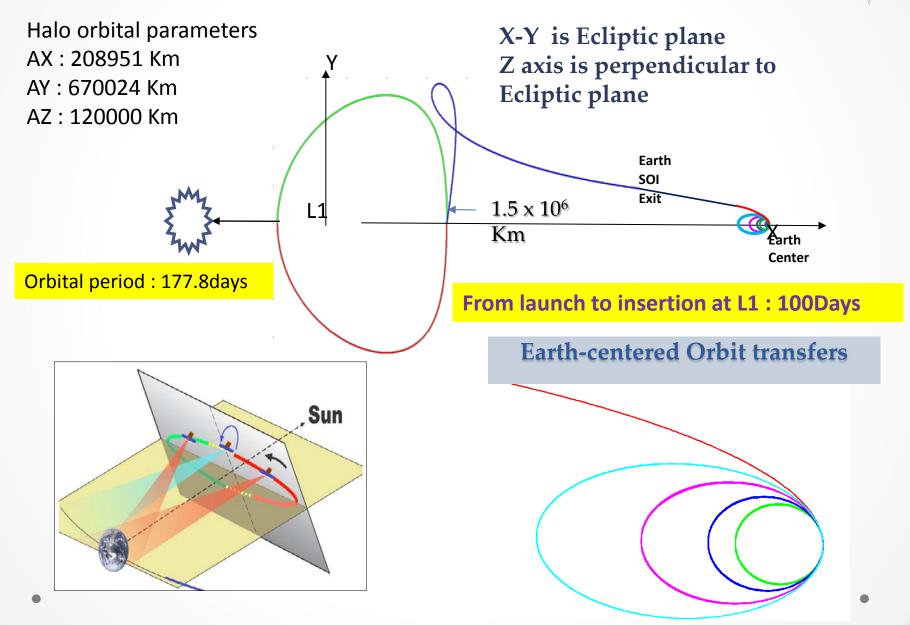
- Realization and launch of a satellite in a Halo orbit around L1.
- Understanding Solar upper atmospheric (chromosphere and corona) dynamics
- Study of chromospheric and coronal heating, Physics of the partially ionized plasma and initiation of the CMEs and flares
- Observe the in-situ particle and plasma environment providing data for the study of particle dynamics from the Sun





Aditya-L1 Mission Profile





Aditya-L1 Payloads

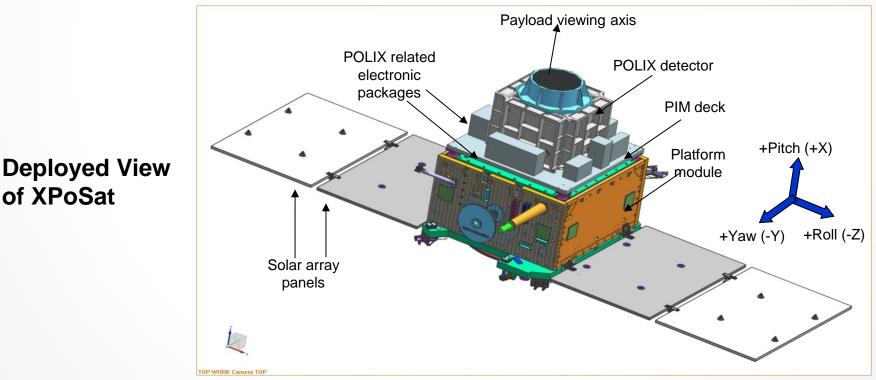


Payloads	Science/Capability
Visible Emission Line Coronagraph (VELC)	Corona/ Imaging , Spectroscopy, Spectro polarimetry
Solar Ultraviolet Imaging Telescope (SUIT)	Photosphere and Chromosphere/ Imaging
Aditya Solar wind Particle Experiment (ASPEX)	Solar wind/ Particle Analyzer In-situ measurement
Plasma Analyser Package For Aditya (PAPA)	Solar wind/ In-situ measurement
Solar Low Energy X-ray Spectrometer (SoLEXS)	Soft X-ray/ In-situ measurement
High Energy L1 Orbiting X-ray Spectrometer (HEL1OS)	Hard X-ray/ In-situ measurement
Advanced Triaxial High Resolution Digital Magnetometers	Measure Magnetic Field/ In-situ measurement

XPoSat Mission Objectives



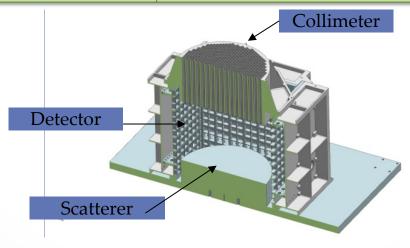
- To measure polarization (degree and direction) of X-ray photons from ~50 potential celestial sources of interest in the energy band of 5-30 keV.
- Mission Life 5 years, Platform Modified IMS-2 Bus
- Payload Polarimeter Instrument in X-rays (POLIX) from Raman Research Institute
- Orbit Circular LEO (500-700 km), Inclination $\leq 30^{\circ}$



XPoSat Payload specifications



Photon collection area	640 cm^2	
Energy range	5-30 keV	
Detectors	Proportional counters	
Total weight	~125 kg	
Overall dimension	~ 650 x 650 x 600 mm ³ (excluding electronics)	
Data generation rate	6 Gbits per day (maximum)	
Scattering element	Beryllium / Lithium	
Rotation (rate)	0.2 rpm	
Pointing accuracy	0.1 deg	



POLIX payload detector inner view

Other Missions being Considered



- Mission to Venus Announcement of Opportunity for experiments to study Venus. Payload capability of 175 kg with 500 W of Power. Initial orbit of around 500 x 60,000 km around Venus.
- Follow on Mars Mission Study team formed to address Science and Technology interests. Mission definition to be formulated.
- Follow on Astrosat Mission Studies in IR, UV and sub-mm wavelengths with next generation optics and detectors.
- **Space Weather Mission** Mission to Sun-Earth Lagrangian point L5 to observe large-scale Solar disturbances that affect earth.

THANK YOU