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**Committee on the Peaceful  
Uses of Outer Space****Information furnished in conformity with the Convention  
on Registration of Objects Launched into Outer Space****Note verbale dated 14 March 2003 from the Permanent Mission of  
Japan to the United Nations (Vienna) addressed to the Secretary-  
General**

The Permanent Mission of Japan to the United Nations (Vienna) presents its compliments to the Secretary-General of the United Nations and, in accordance with article IV of the Convention on Registration of Objects Launched into Outer Space (General Assembly resolution 3235 (XXIX), annex), has the honour to transmit information concerning the launching of the Japanese satellites JCSAT-8 and N-STAR c, the Data Relay Test Satellite "Kodama", the Advanced Earth Observation Satellite "Midori-II", the Whale Ecology Observation Satellite "Kantakun", Micro LabSat and the Unmanned Space Experiment Recovery System spacecraft (see annex).



## Annex

### Registration data for Japanese space launches\*

#### A. JCSAT-8

1. Name of flight object: JCSAT-8
2. Designation: 2002-015A
3. Name of launching State: Japan
4. Date and time of launch: 29 March 2002 at 0129 UT
5. Location of launch: Guiana Space Centre, Kourou, French Guiana
6. Basic orbital parameters: (as at 9 May 2002)
  - (a) Nodal period: 1,436 minutes
  - (b) Inclination: 0.012 degrees
  - (c) Apogee: 35,797 kilometres
  - (d) Perigee: 35,778 kilometres
7. General function: Domestic and international telecommunications and domestic broadcasting
8. Launch vehicle: Ariane 44L
9. Launching organization: Arianespace

#### B. N-STAR c

1. Name of flight object: N-STAR c
2. Designation: 2002-035B
3. Name of launching State: Japan
4. Date and time of launch: 5 July 2002 at 2321 UT
5. Location of launch: Guiana Space Centre, Kourou, French Guiana
6. Basic orbital parameters: (as at 4 August 2002)
  - (a) Nodal period: 1,436 minutes
  - (b) Inclination: 0.06 degrees
  - (c) Apogee: 35,790 kilometres
  - (d) Perigee: 35,772 kilometres
7. General function: Domestic telecommunications

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\* The registration data are reproduced in the form in which they were received.

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8. Launch vehicle: Ariane 5
  9. Launching organization: Arianespace

### **C. Data Relay Test Satellite “Kodama”**

1. Name of flight object: Data Relay Test Satellite (DRTS) “Kodama”
2. Designation: 2002-042B
3. Name of launching State: Japan
4. Date and time of launch: 10 September 2002 at 0820 UT
5. Location of launch: Tanegashima Space Centre, Kagoshima, Japan
6. Basic orbital parameters: (as at 11 October 2002)
  - (a) Nodal period: 1,436 minutes
  - (b) Inclination: 0.07 degrees
  - (c) Apogee: 35,798.6 kilometres
  - (d) Perigee: 35,774.2 kilometres
7. General function: The main objective of DRTS is the conducting of inter-satellite communication experiments to relay data between the target spacecraft (low-Earth-orbit satellite, space stations and so forth) and ground stations. The satellite is located over 90.75 degrees East.
8. Launch vehicle: H-II A Launch Vehicle F3 (H-IIAF3)
9. Launching organization: National Space Development Agency of Japan

### **D. Advanced Earth Observation Satellite “Midori-II”**

1. Name of flight object: Advanced Earth Observation Satellite II (ADEOS II) “Midori-II”
2. Designation: 2002-056A
3. Name of launching State: Japan
4. Date and time of launch: 14 December 2002 at 0131 UT
5. Location of launch: Tanegashima Space Centre, Kagoshima, Japan
6. Basic orbital parameters: (as at 19 December 2002)
  - (a) Nodal period: 101 minutes
  - (b) Inclination: 98.7 degrees
  - (c) Apogee: 820 kilometres
  - (d) Perigee: 803 kilometres

7. General function: ADEOS II has been developed with the aim of advancing the Earth-observing technologies. ADEOS II is the successor to the first Advanced Earth Observation Satellite, whose mission was to acquire data on the mechanism of global environmental change, including global warming, and on weather phenomena and so forth.
8. Launch vehicle: H-II A Launch Vehicle F4 (H-IIAF4)
9. Launching organization: National Space Development Agency of Japan

### **E. Whale Ecology Observation Satellite “Kantakun”**

1. Name of flight object: Whale Ecology Observation Satellite (WEOS) “Kantakun”
2. Designation: 2002-056C
3. Name of launching State: Japan
4. Date and time of launch: 14 December 2002 at 0131 UT
5. Location of launch: Tanegashima Space Centre, Kagoshima, Japan
6. Basic orbital parameters: (as at 17 December 2002)
  - (a) Nodal period: 101 minutes
  - (b) Inclination: 98 degrees
  - (c) Apogee: 812 kilometres
  - (d) Perigee: 774 kilometres
7. General function: The WEOS system consists of a small satellite in a polar orbit, probes attached to whales and a ground station for satellite data reception. The objective of the system is to obtain data on whale ecology and the ocean environment.
8. Launch vehicle: H-II A Launch Vehicle F4 (H-IIAF4)
9. Launching organization: National Space Development Agency of Japan

### **F. Micro LabSat**

1. Name of flight object: Micro LabSat ( $\mu$ -LabSat)
2. Designation: 2002-056D
3. Name of launching State: Japan
4. Date and time of launch: 14 December 2002 at 0131 UT
5. Location of launch: Tanegashima Space Centre, Kagoshima, Japan
6. Basic orbital parameters: (as at 15 December 2002)

- (a) Nodal period: 101 minutes
- (b) Inclination: 98.7 degrees
- (c) Apogee: 811 kilometres
- (d) Perigee: 767 kilometres
- 7. General function: Bus system feasibility for 50 kg-class small satellite; new separator feasibility; and remote inspection technology.
- 8. Launch vehicle: H-II A Launch Vehicle F4 (H-IIAF4)
- 9. Launching organization: National Space Development Agency of Japan

### **G. Unmanned Space Experiment Recovery System spacecraft**

- 1. Name of flight object: Unmanned Space Experiment Recovery System (USERS) spacecraft
- 2. Designation: 2002-042A
- 3. Name of launching State: Japan
- 4. Date and time of launch: 10 September 2002 at 0820 UT
- 5. Location of launch: Tanegashima Space Centre, Kagoshima, Japan
- 6. Basic orbital parameters: (as at 8 October 2002)
  - (a) Nodal period: 95 minutes
  - (b) Inclination: 30.4 degrees
  - (c) Apogee: 515 kilometres
  - (d) Perigee: 501 kilometres
- 7. General function: The mission of the USERS spacecraft is:
  - (a) To establish self-re-entry and return the Unmanned Space Experiment Recovery System;
  - (b) To process high-temperature superconducting material under microgravity conditions in orbit;
  - (c) To verify commercial parts in a space environment.
- 8. Launch vehicle: H-II A Launch Vehicle F3 (H-IIAF3)
- 9. Launching organization: National Space Development Agency of Japan

10. Remark: The re-entry module, part of the USERS spacecraft, is scheduled to separate, re-enter the atmosphere and be guided to a recovery area in the open sea off the Ogasawara islands during the period following May 2003.
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