Day #2 01.21 21:00~23:00 (JST)

KiboCUBE Academy

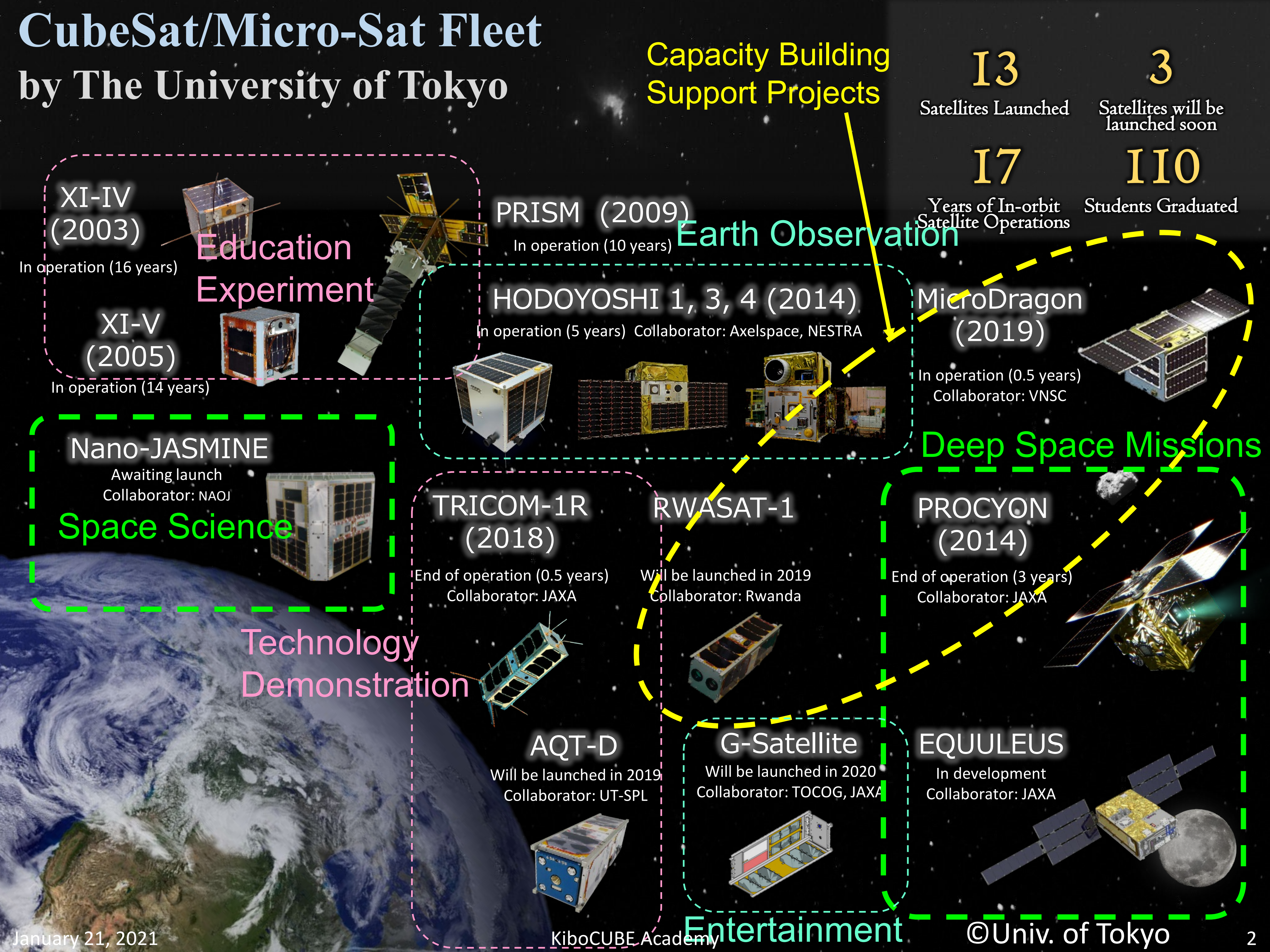
Lecture 2-1

Overview of Satellite Development Process

University of Tokyo

Shinichi Nakasuka

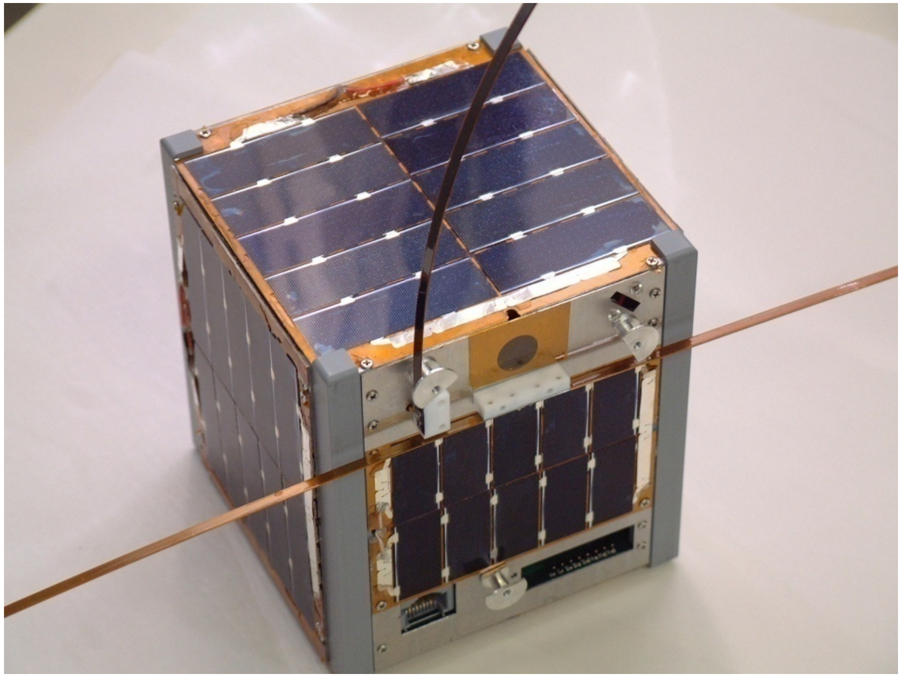
This lecture is NOT specifically about KiboCUBE and covers GENERAL engineering topics of space development and utilization for CubeSats.
The specific information and requirements for applying to KiboCUBE can be found at:
https://www.unoosa.org/oosa/en/ourwork/psa/hsti/kibocube.html


CubeSat/Micro-Sat Fleet
by The University of Tokyo
Capacity Building
Support Projects
13
Satellites Launched
3
Satellites will be launched soon
17
Years of In-orbit Satellite Operations
110
Students Graduated
XI-IV
(2003)
In operation (16 years)
Education
Experiment
XI-V
(2005)
In operation (14 years)
PRISM (2009)
In operation (10 years)
Earth Observation
HODOYOSHI 1, 3, 4 (2014)
In operation (5 years) Collaborator: Axelspace, NESTRA
MicroDragon
(2019)
In operation (0.5 years)
Collaborator: VNSC
Deep Space Missions
Nano-JASMINE
Awaiting launch
Collaborator: NAOJ
Space Science
TRICOM-1R
(2018)
End of operation (0.5 years)
Collaborator: JAXA
RWASAT-1
Will be launched in 2019
Collaborator: Rwanda
PROCYON
(2014)
End of operation (3 years)
Collaborator: JAXA
Technology
Demonstration
AQT-D
Will be launched in 2019
Collaborator: UT-SPL
G-Satellite
Will be launched in 2020
Collaborator: TOCOG, JAXA
EQUULEUS
In development
Collaborator: JAXA
Entertainment
January 21, 2021
KiboCUBE Academy
©Univ. of Tokyo
2

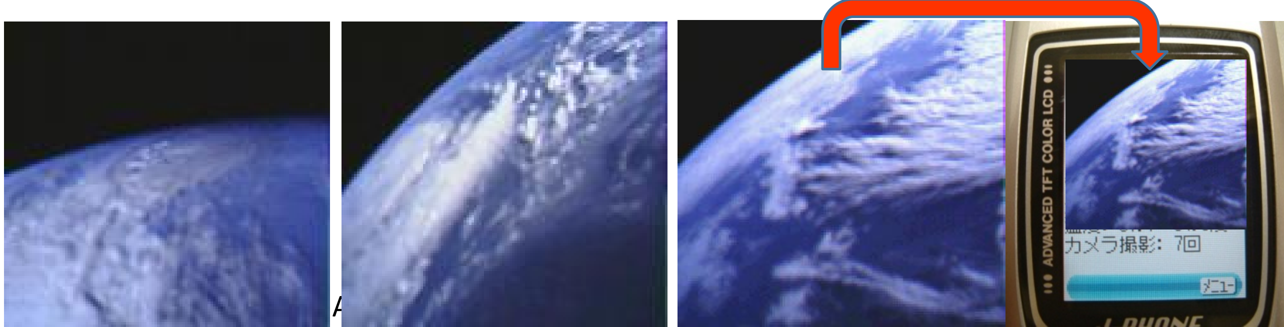
Example – 1U CubeSat “XI-IV”

Mission: Pico-bus technology demonstration in space, Camera experiment
Developer: University of Tokyo
Launch: ROCKOT (June 30, 2003) in Multiple Payload Piggyback Launch

Size	10x10x10[cm] CubeSat
Weight	1 [kg]
Attitude control	Passive stabilization with permanent magnet and damper
OBC	PIC16F877 x 3
Communi-cation	VHF/UHF (max 1200bps) (amateur frequency band)
Power	Si solar cells for 1.1 W
Camera	640 x 480 CMOS
Life time	Already over 17 years



Captured Earth images are distributed to mobile phones



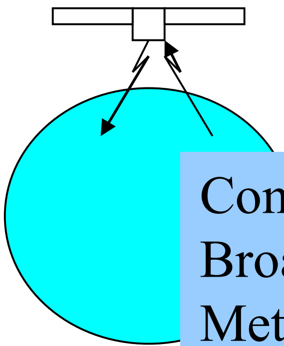
Contents

1. Create good mission!

1. Why we use space ?

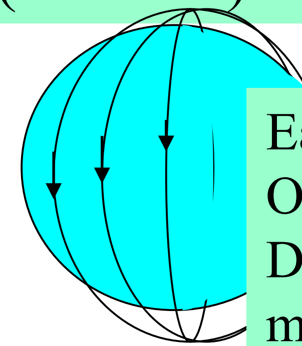
Features of space which lead to space utilization

①Providing 3D View



Communication, Broadcasting, GPS, Meteorological, Earth Observation

②High Speed Earth Coverage (5-8km/s)



Earth Observation, Disaster monitoring

③Above Atmosphere

- Space Telescope, Various spectral observations,
- Solar power generation

④Long time μG environment

- New material/medicine
- Life science experiment

⑤Space as Exploration Target

- Observation of Planets, Small bodies
- Particles, Fields, etc.

⑥"Human in Space" (travel)

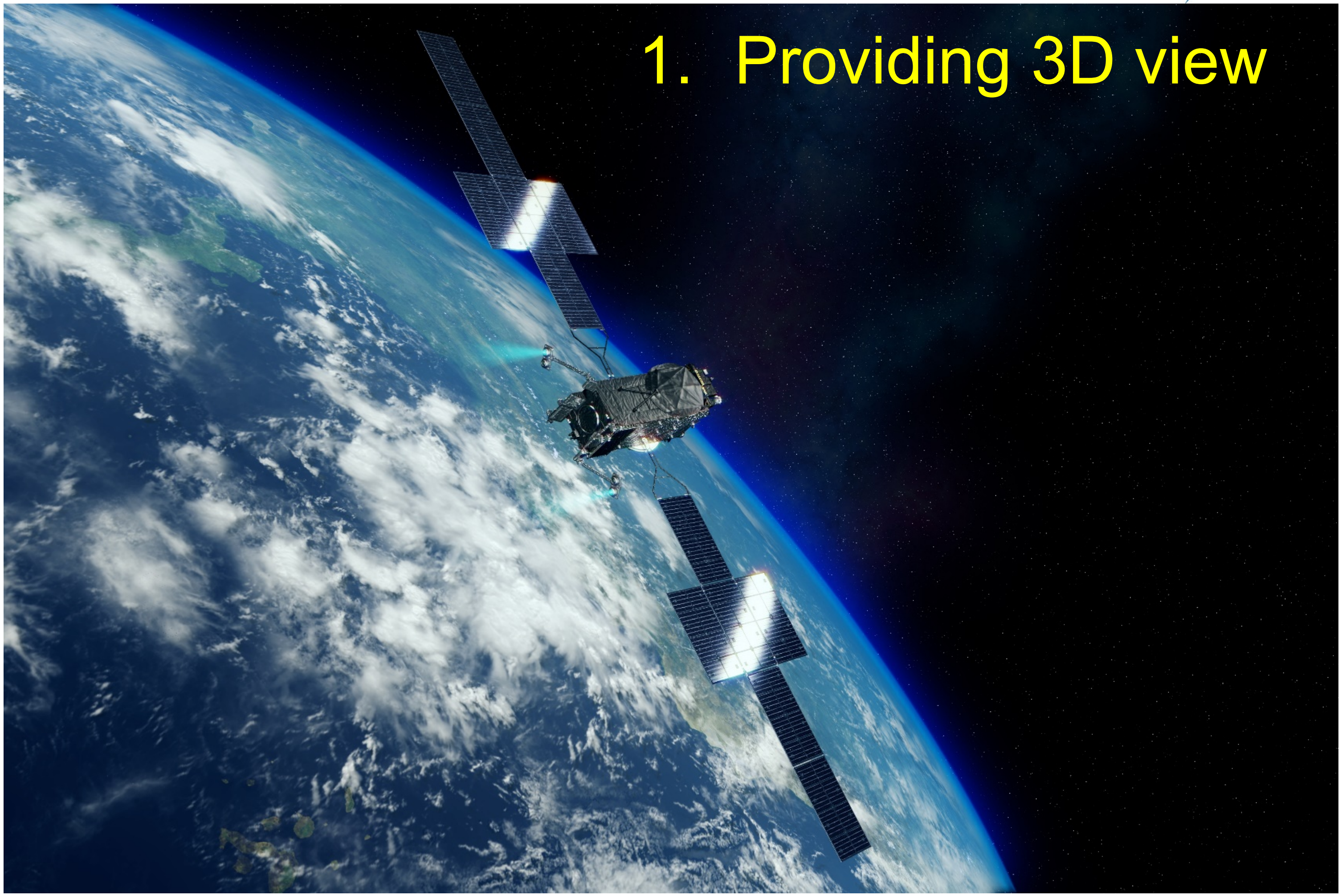
⑦⑧---- Waiting for other new ideas !

Practice 1

- Which special features (from 1-6) are utilized in the satellites in following slides?

1. Providing 3D view
2. High speed coverage of the Earth
3. Staying above atmosphere
4. Obtaining long time micro-g environment
5. Space as exploration target
6. Human existence in space
7. Others

Communication/broadcasting satellite

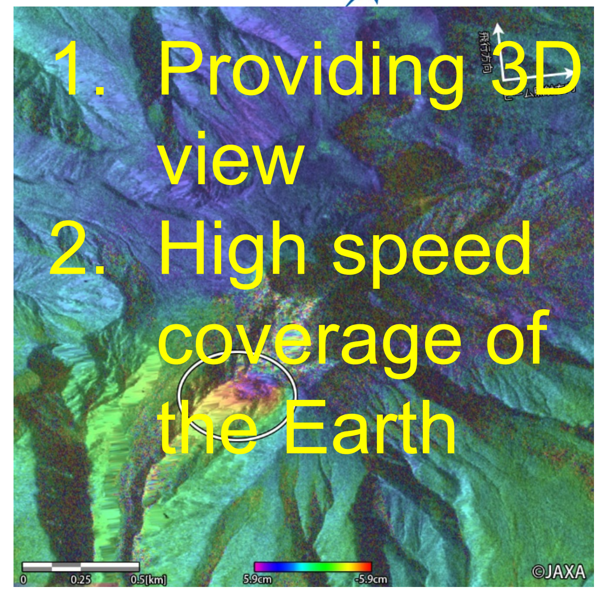


ETS-9 (Engineering Test satellite)

Synthetic Aperture Radar (SAR)

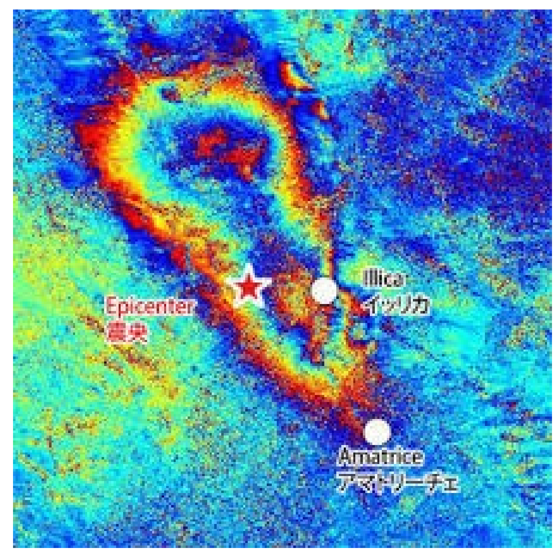
ALOS-2

©JAXA

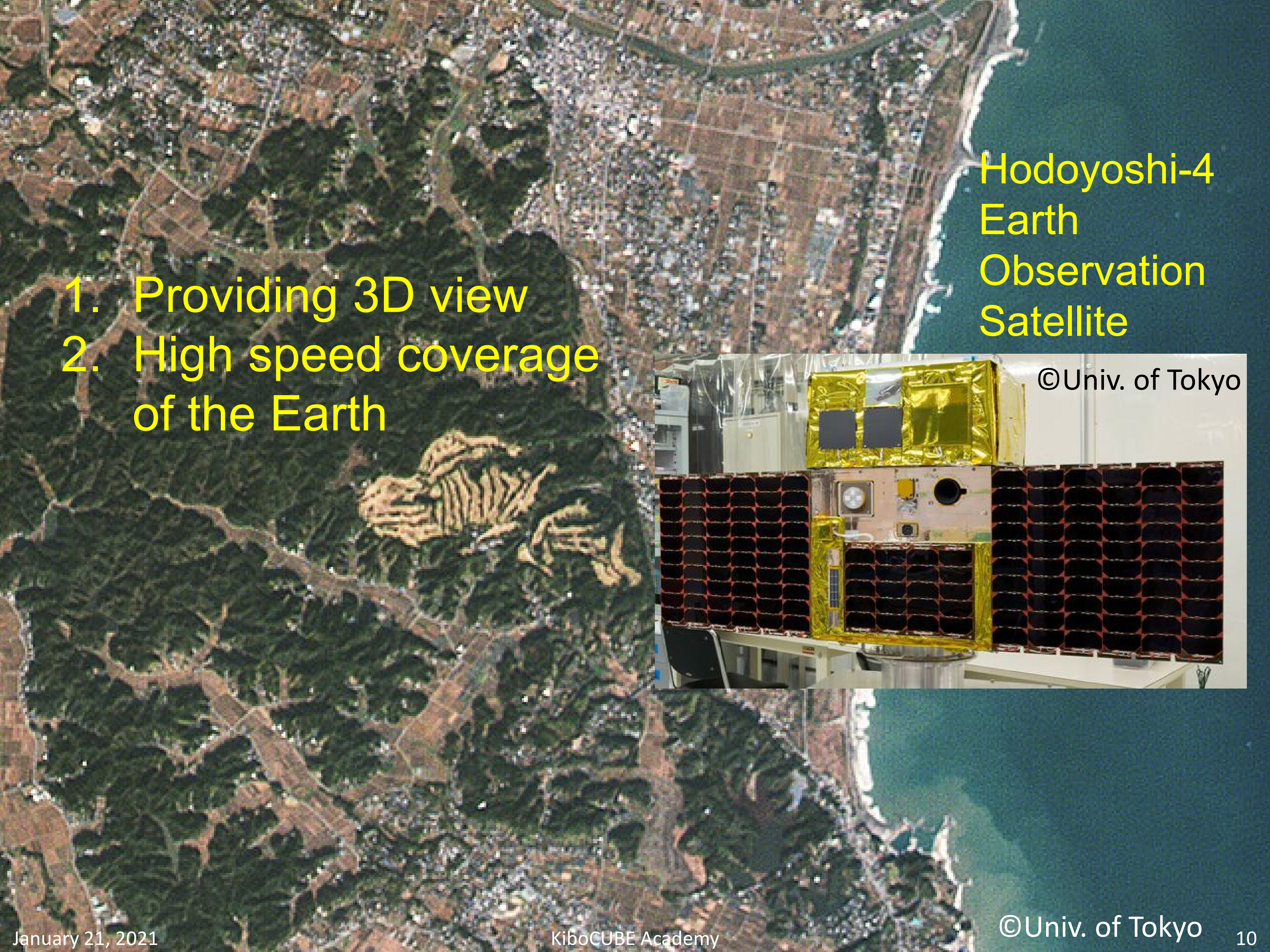


1. Providing 3D view
2. High speed coverage of the Earth

©JAXA



©JAXA



Hodoyoshi-4
Earth
Observation
Satellite

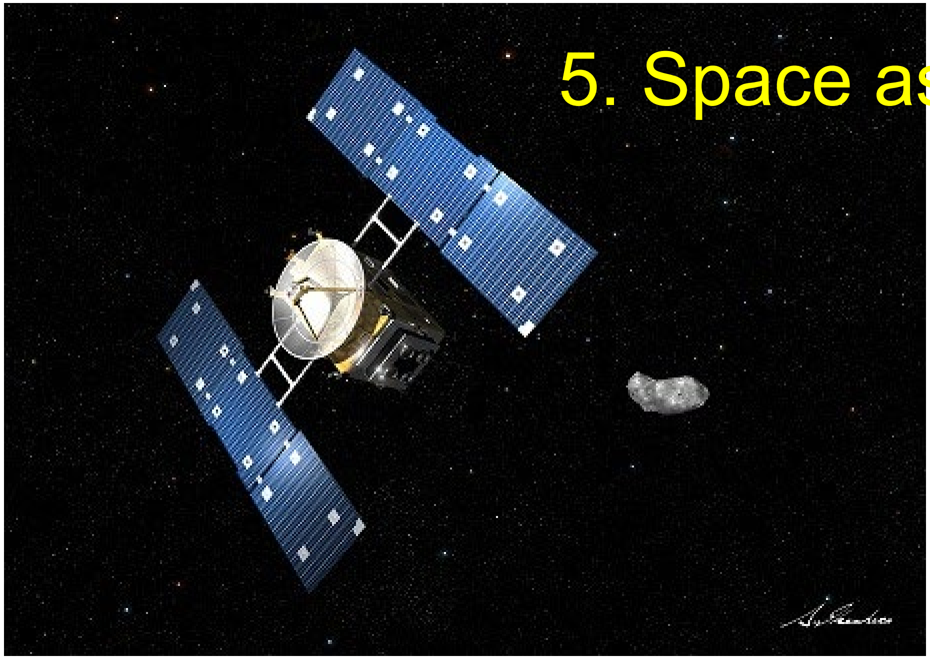
1. Providing 3D view
2. High speed coverage of the Earth

Hubble: Space Telescope

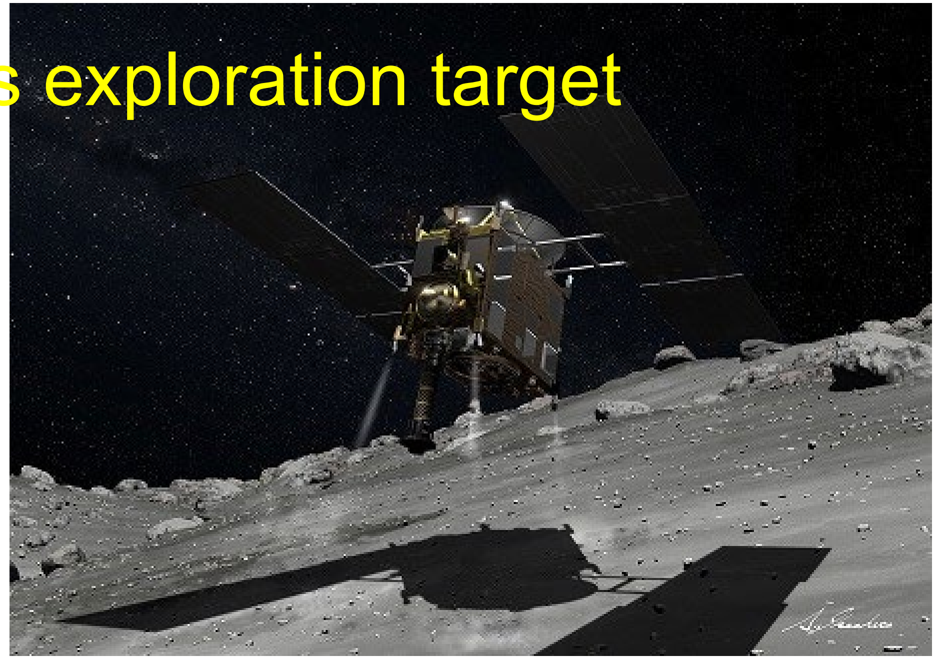
Hubble Space Telescope (1990～)

Deep Space Exploration Probe

5. Space as exploration target



©JAXA



©JAXA

"HAYABUSA（MUSES-C）"

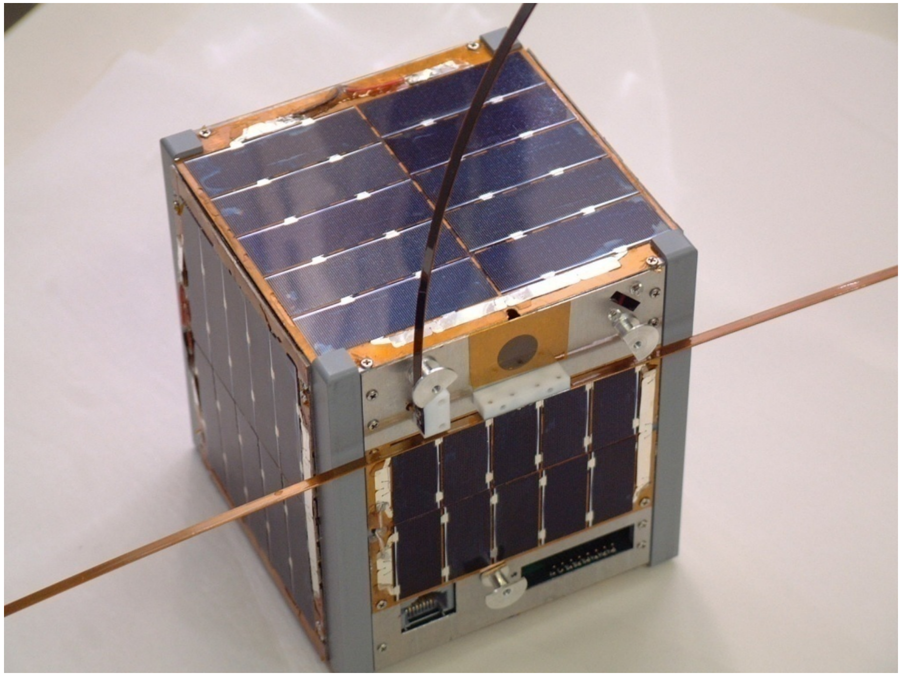
Launched in 2003, Returned 2010

"Sample return" from asteroid "Itokawa"

Lunar Exploration Satellite

Kaguya

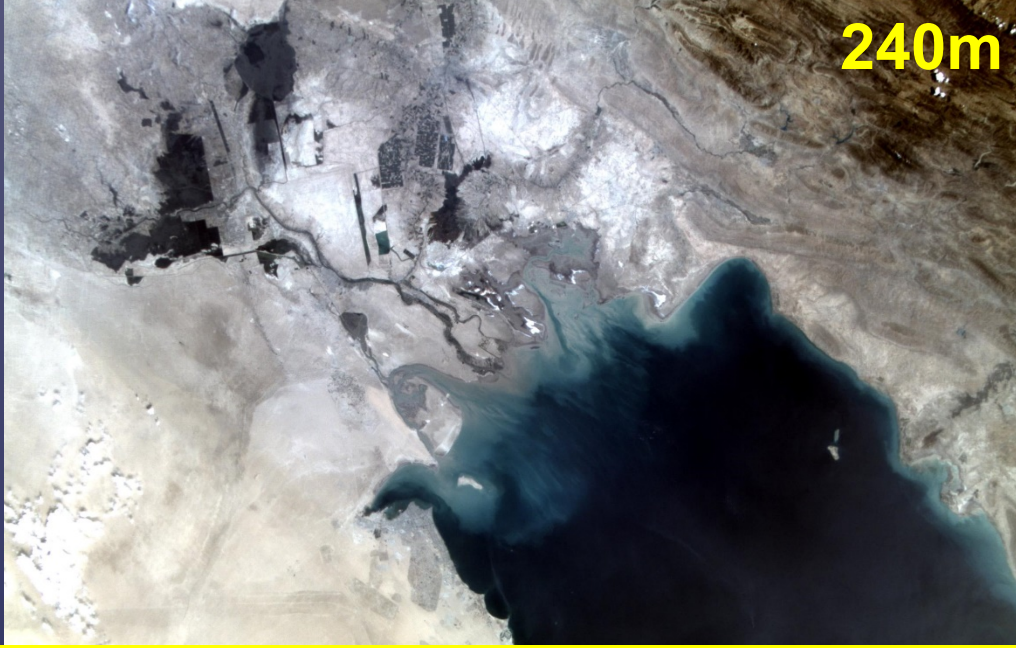
What kind of mission possible for CubeSat ?



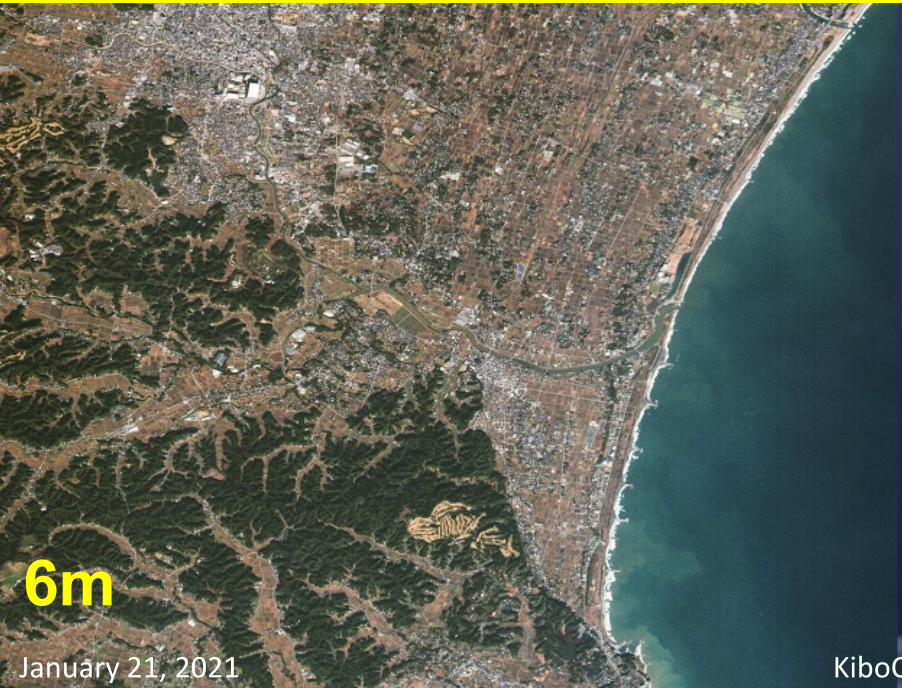
Size: 10cm x 10cm x 10cm < around 1kg
Orbit: near ISS orbit if deployed from ISS

Example Missions for CubeSat (1)

- Earth observation with onboard camera
  1. What is your observation target ?
     - Agricultural field, disaster, land usage, sea shore, houses…..
  2. How much resolution is required ? (5m, 10m, 100m ?)
     - Limitation: resolution is strictly limited by small aperture size
  3. Wavelength: red, green, blue, red edge, near infrared…
- Star observation with onboard camera
  - Limitation: stars are not so bright and long exposure time is necessary, which requires high attitude stability for a long time
- Other sensing from space or in-situ observation
  - Special sensors are required: collaboration with researchers of the earth, atmosphere, space science, etc.
  - In-situ plasma, particle, magnetic field, atmospheric drag..



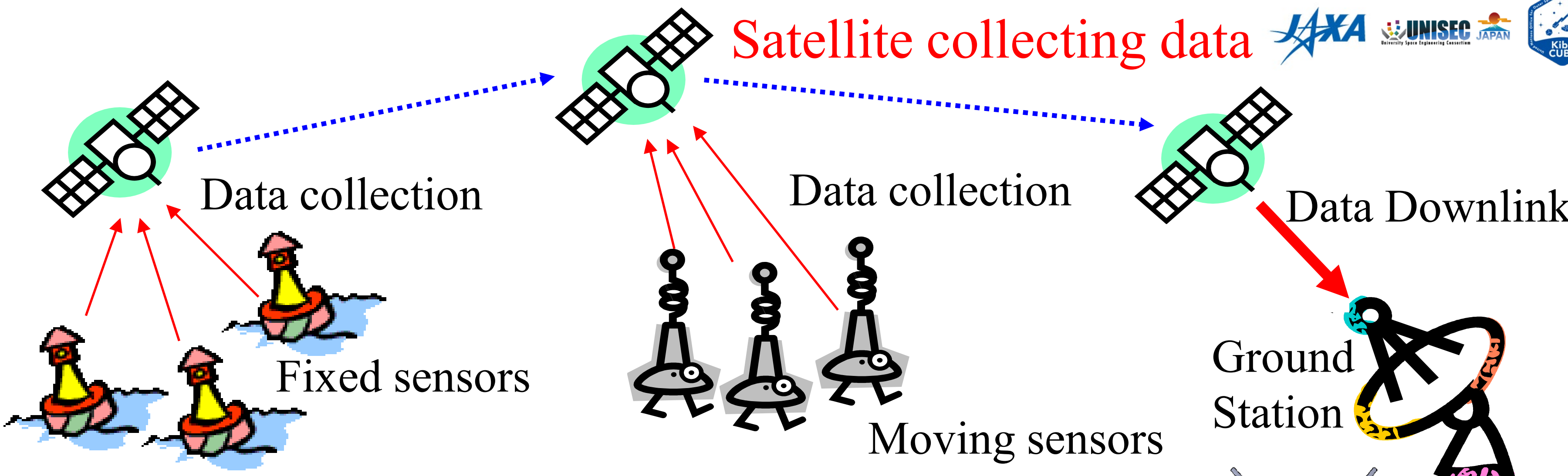
What spatial resolution do you need for your mission ?



Example Missions for CubeSat (2)

- Communication with ground
  - From ground to satellite
    - Risk information, personal data, health data, disaster related information. (effective where there is no mobile phone network)
    - Limitation: data rate is limited by power, and antenna size
  - From satellite to ground
    - Broadcasting messages such as warning signals, education, etc.
  - Message relay: uplink, carry and download message
- Experiment inside CubeSat
  - Experiment in micro-gravity, space environment, etc.
    - How to automatically do experiment, record and downlink to ground?
- Technological experiments or demonstrations
- Others

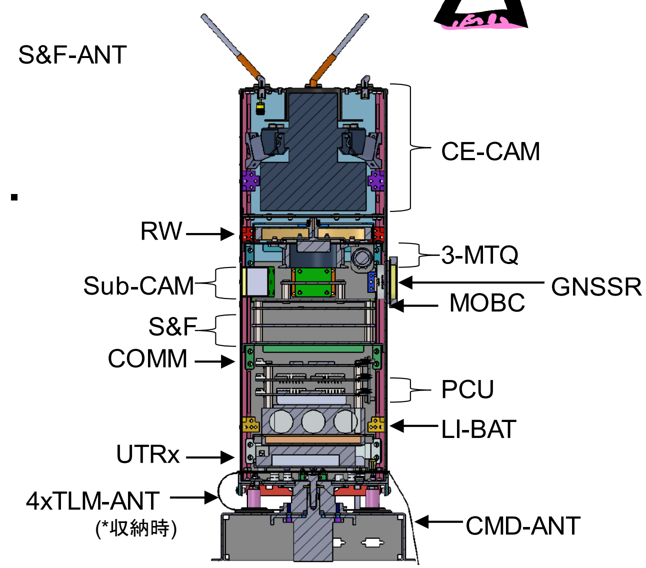
Example) IoT Mission



Application areas: disaster prediction, water level monitoring, soil moisture, PH…..

Key Issue: **<u>How to send data with very low RF power to the satellite ?</u>**

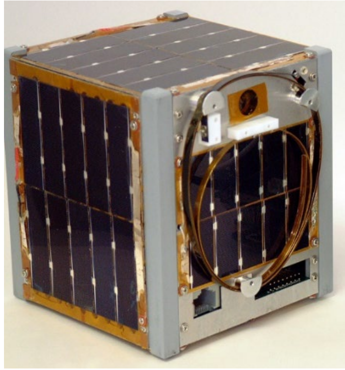
8 - 20mW RF power, low data rate (300bps) transmission was successful.

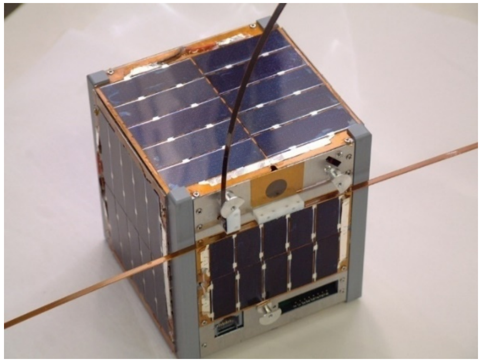


3kg TRICOM-1R

Important Tactics for Mission Creation

- Do not aim at high level mission
  - Aperture size and focal length limit spatial resolution
  - Uplink/Downlink communication speed is limited
- Should be installed into 10cm cubic shape before release from the rocket
  - Antenna and other appendages should be initially stowed and deployed after release from rocket
  - Generated power is around 1 - 2W on average (1U)
- "Satellite bus system functioning" in itself is a good mission if it is your first project





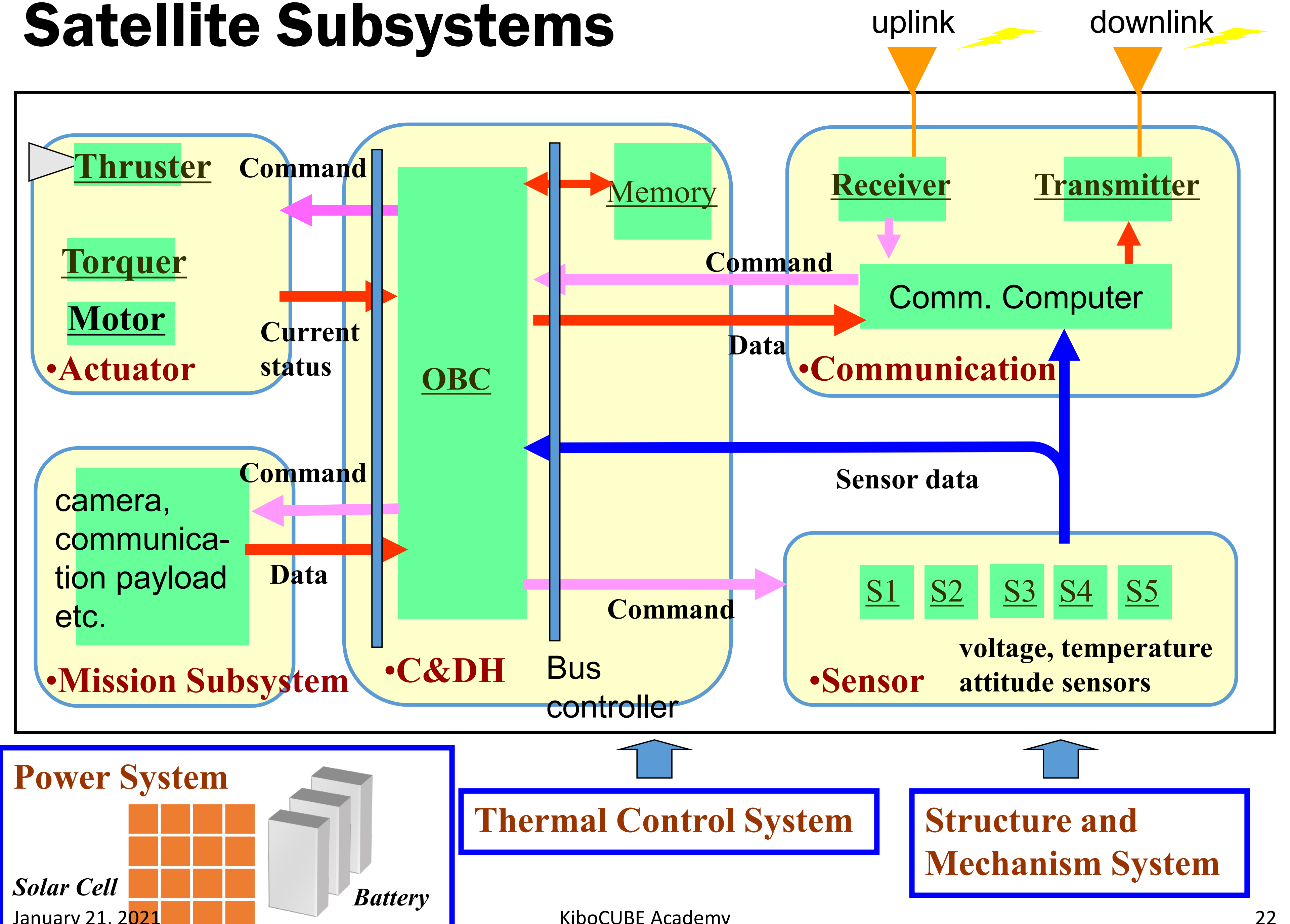
Example) CubeSat “XI-IV” Missions

- Realizing satellite bus functions for a certain period
  - Functioning of satellite as designed
  - Making it survive in space environment (for minimum 1 year)
  - Communication using amateur radio frequency
- Capturing and downlinking Earth photos
  - XI-IV captured as many photos as possible and only photos which have Earth images were stored and downlinked

  - 1200bps requires 3 - 4 days to downlink 1 photo

2. System Level Design

Satellite Subsystems



Breakdown of mission requirements

- Attitude control subsystem
  - Free motion
  - Gravity gradient, passive magnetic control
  - Three axis control: on-board antenna or camera is to be directed towards the some location on earth
    - accuracy of attitude determination, orientation control and stability
- Communication subsystem
  - How much data should be downlinked to ground per day decides downlink communication speed
- Power subsystem
  - Maximum and average power requirement should be satisfied
- Thermal control subsystem
  - Temperature control for each component to be in allowable range
- Structure and mechanism subsystem
  - Should endure launch load, with deployment mechanism if required

System Level Design: "Budgeting"

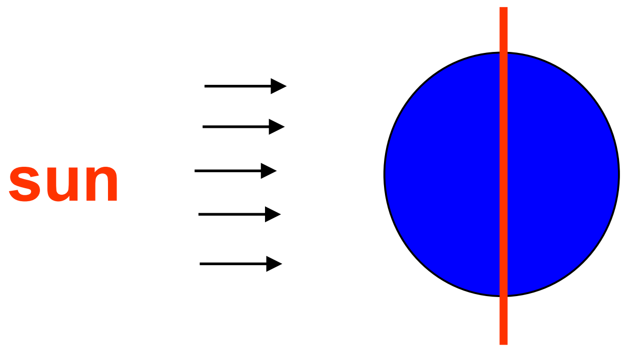
- Input to weight/size/power budgeting -

Equipment	weight (g)	Size (mm)	Power (mW)	Tempera-ture (℃)
Transmitter	90	80x80x10	800	-10 ～ 50
Receiver	80	80x80x10	100	-10 ～ 50
Antenna	30	80x80x8	500	-10 ～ 50
Computer	100	80x80x10	200	-10 ～ 50
Power Dist.	120	40x40x10	50	-10 ～ 50
Battery	150	90x90x10	-	-5 ～ 40
Solar cells	80	50x30x1	-	-20 ～ 100
Wheels	100	20x20x30	500	-10 ～ 50
Mag. sensor	100	20x20x10	200	-10 ～ 50
Mag.torquer	150	Φ10x50	1000	0 ～ 30

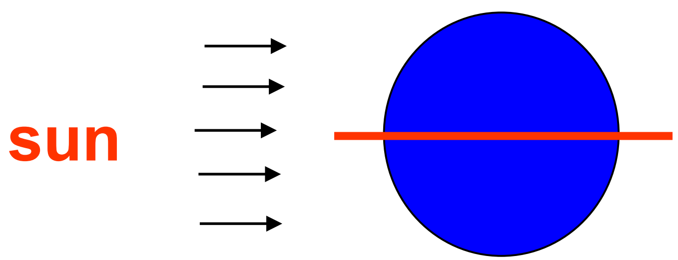
Power Budgeting (1)

Calculate power generation considering orbit

- Estimate eclipse percentage from the relationship between orbit and sun direction
  - In eclipse period, battery is used to supply power
- Calculate average power generation based on orbit information, attitude control strategy and solar cell implementation design



Almost 0 % eclipse

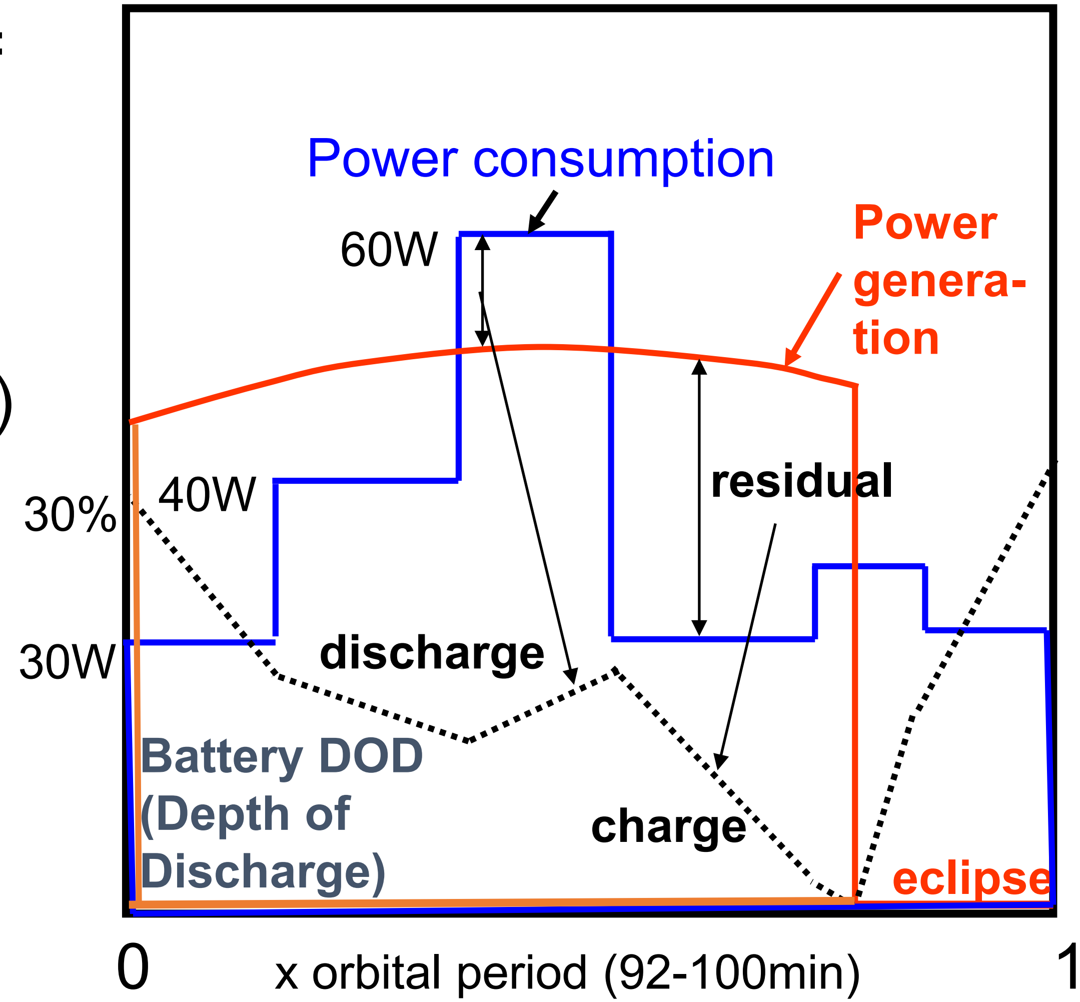


40 – 48 % eclipse

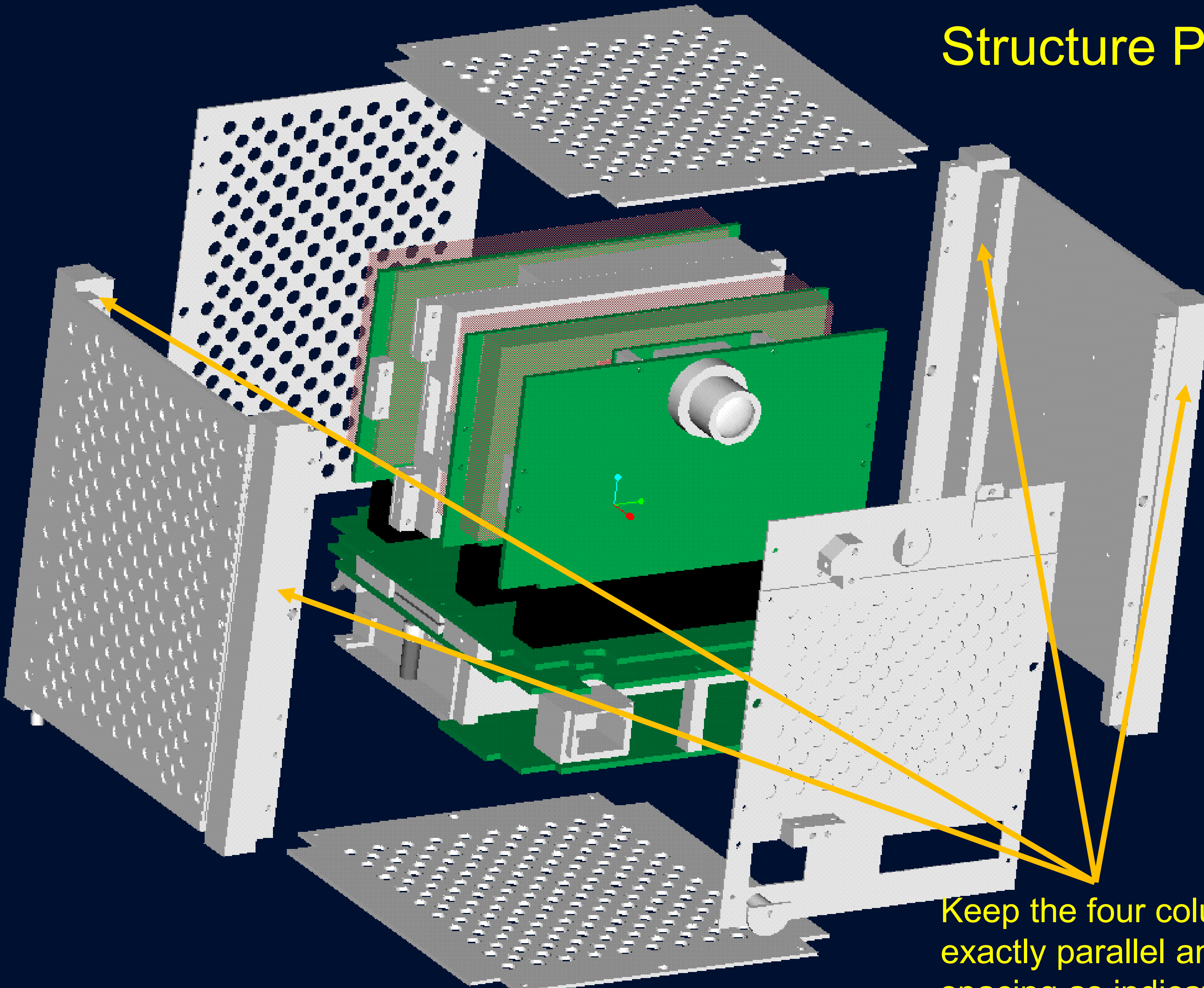
Power Budgeting (2)

Balance power generation and consumption

- Power consumption = power generation in one orbital period
  - Solar cells sizing ($1358 W/m^2$, 16～28 %)
- Battery to provide additional power
  - Battery sizing to make max DOD about 30 %



Structure Plan



Keep the four columns exactly parallel and spacing as indicated

3. Subsystems and Teaming

Satellite Subsystems

- Mission subsystem
- Command & Data Handling (C&DH)
- Communication (COM)
- Electric Power (EPS)
- Attitude Determination and Control (ADCS)
- Structure and Mechanism
- Thermal Control
- (Special subsystems required for your mission)

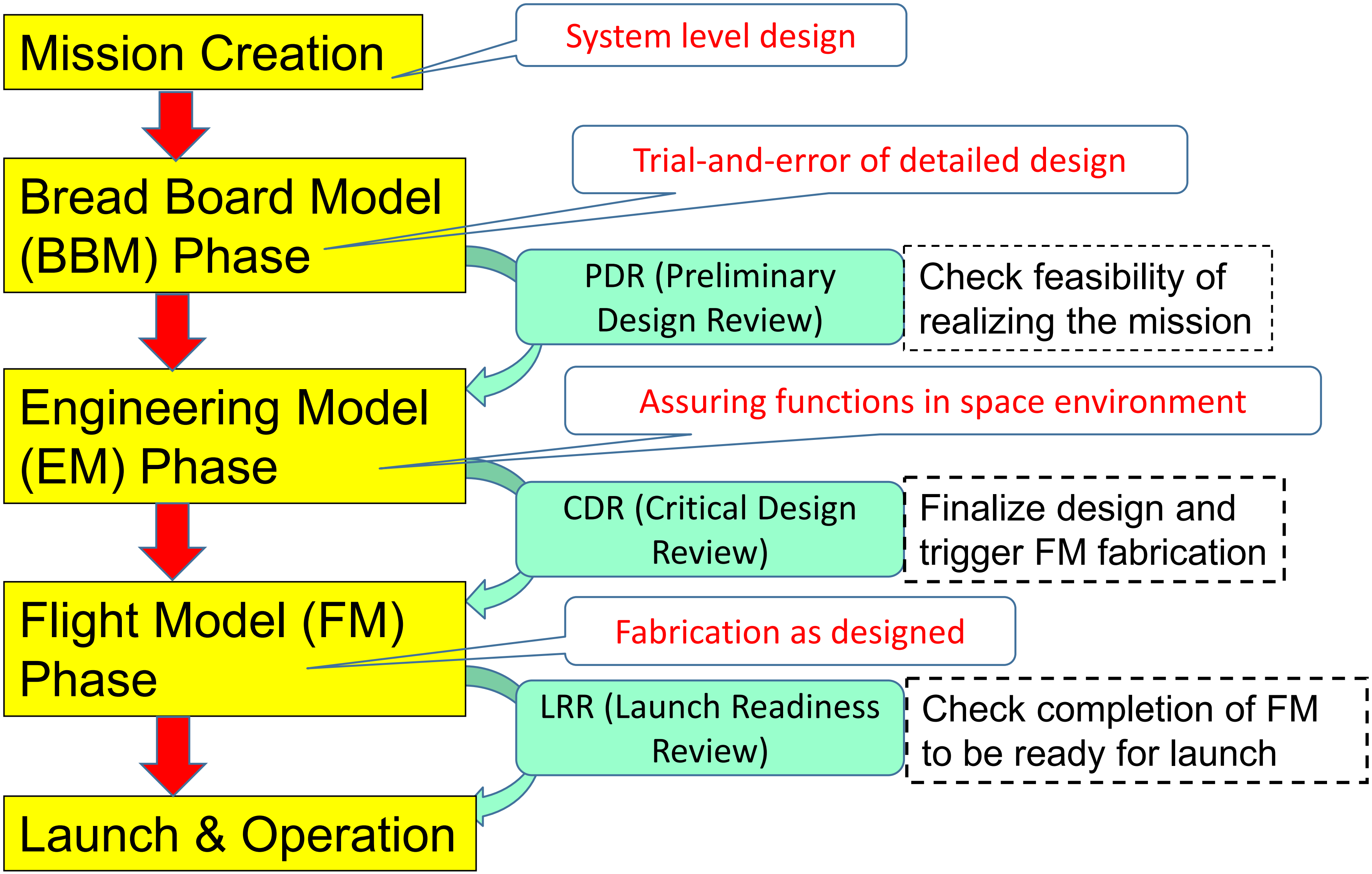
- Ground Station

Teaming

- Based on subsystems
  - Ex) assigning two subsystems to each member
- Based on administrative roles:
  - Project Manager (PM), Sub-manager (Sub-PM)
  - Budget management
  - Parts/components search and purchase
  - Documentation and data control (Web, ICD….)
  - External relationships & promotion
    (getting permission, regulations, seeking funds, etc.)
  - Consultation Team (experts on electronics, communication, structure, ground test, etc.)

4. Flow of Satellite Development and Review Meetings

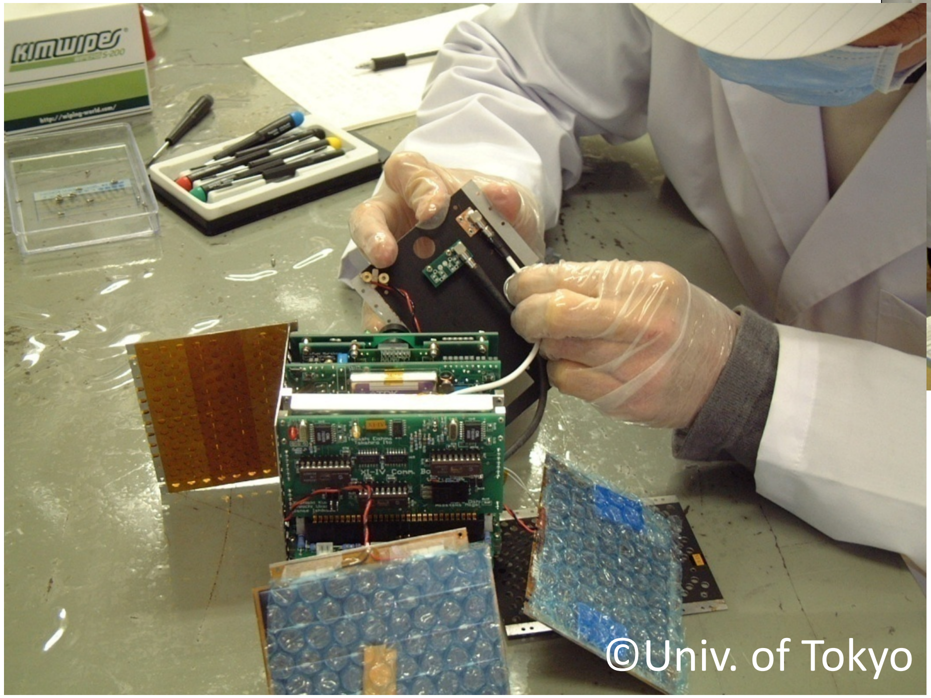
Water Flow Project Management



“Table-sat” assures proper functionalities

FM Integration in Clean Room

Mounting solar cells on flight model in clean booth



Final Integration Procedure

5. Ground Tests and Feedbacks

Usually Required Ground Tests

Mission Creation

System level design

Bread Board Model (BBM) Phase

Trial-and-error of detailed design

Radiation tests
Simple function tests
(table sat)

Engineering Model (EM) Phase

Assuring functions in space environment

Space environment tests (vibration, shock, thermal, thermal vacuum tests) with strong load (Qualification Test)

Flight Model (FM) Phase

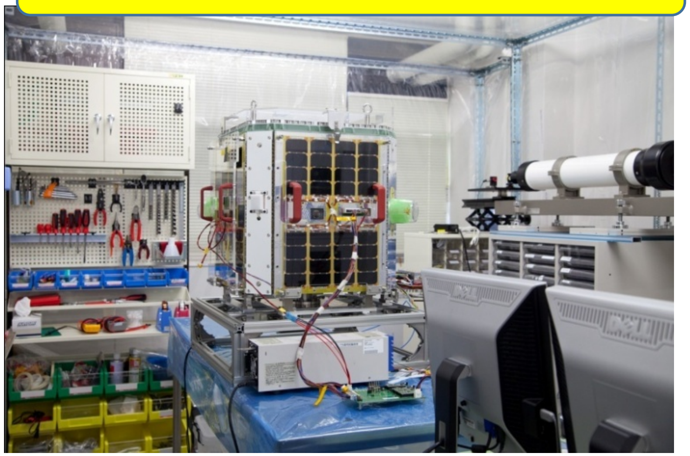
Fabrication as designed

Space environment tests (vibration, shock, thermal, thermal vacuum tests)
with launch load (Acceptance Test)

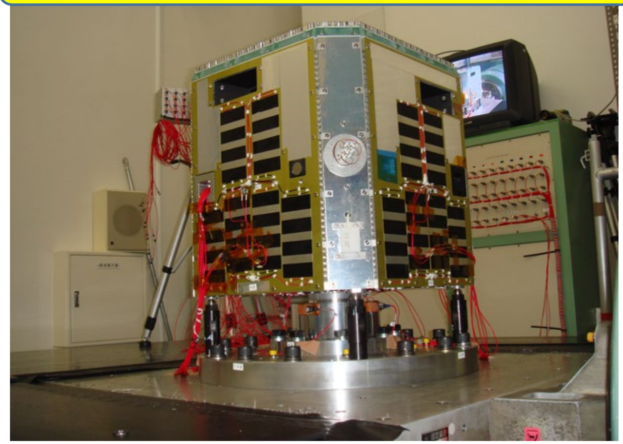
Launch & Operation

Satellite Development & Operation Facilities

Clean Booth (class100,000)



Vibration Table (25g rms)



Thermal bath
(-70 ～100 ℃)

Thermal Vacuum Chamber

UHF/VHF/S-band Ground Station Antenna

- Solar Simulator
- RF test room
- Vacuum Chamber

Find external facilities !

Example) CubeSat "XI-IV" Development

1.8 years of Development

2000

8 9 10 11 12

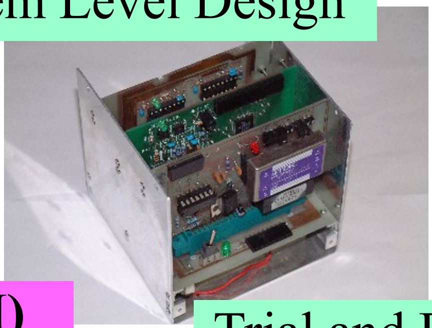
2001

1 2 3 4 5

Mission Creation

System Level Design

Radiation Table-sat Test



XI-I（BBM)

Trial and Error

Table Sat Test

PDR

XI-II（BBM)

6 7 8 9 10 11 12

2002

1 2 3

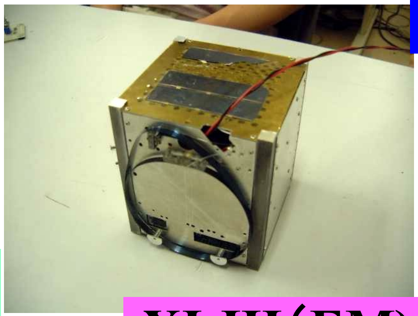
CDR

LRR

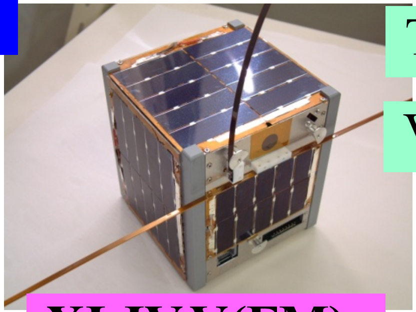
Long Range Comm. Test

Vibration

Thermal Vacuum



XI-III（EM)



XI-IV,V(FM)

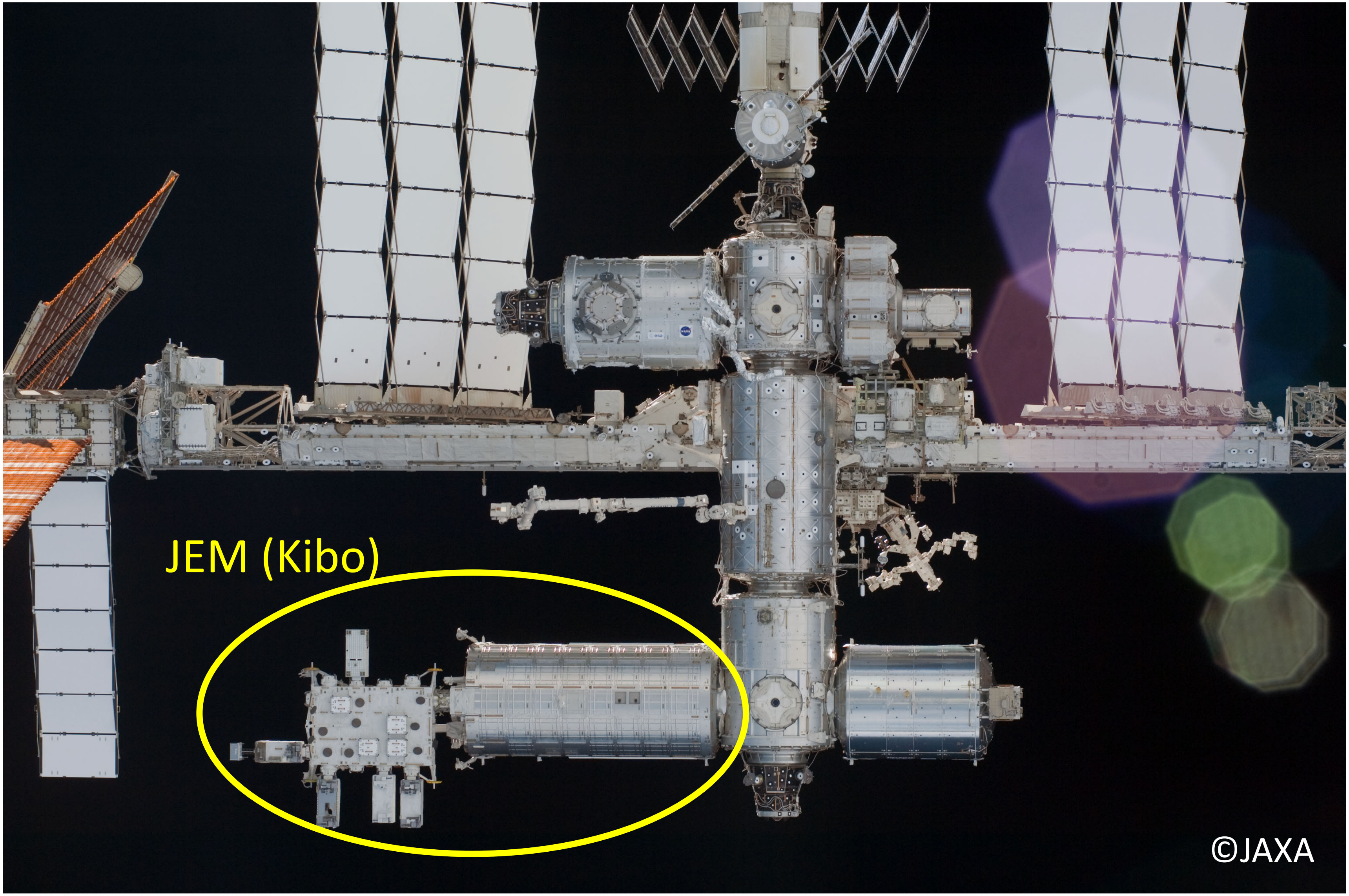
Thermal

Vacuum

Operation training

6. Launch Arrangement and Interface Control

Deployment from ISS



S130E012215

Considerations in Launch Interface Control

- CubeSats will be deployed from rockets or ISS, and POD is used for interface (for ISS case J-SSOD)"
- The fabricated CubeSat shape and size should perfectly match with the POD internal size
  - The specifications of POD will be informed by launch provider (for ISS case, JAXA)
  - The CubeSat's four vertical columns should be parallel and have exactly the same spacing as required
- "Cold Launch": CubeSat is switched off during the launch and should be switched on automatically after release from POD

Safety Requirements for Deployment from ISS

Your CubeSat should follow various rules.

- Structure strength and rigidity
  - Tolerant against acceleration and vibration load, etc.
  - Fundamental frequency has to be over 60 Hz
- Size and weight
  - Should match the J-SSOD size and weight requirement
- May have to wait for long time before launch
  - Waiting time may be as long as several months to one year
- Environmental conditions
  - Should be tolerant against given pressure change, temperature and humidity environment, etc.
- Many other rules will be instructed by launch provider
  - You should prove them by simulations, ground tests and/or fit check etc.
- Please refer to JEM Payload Accommodation Handbook

7. Ground Station and Satellite Operation

Find ground station !

Preparations for Ground Operation

- Frequency permission has to be obtained before release from ISS or rocket, which requires more than 1.5 year negotiation with your authority of communication, and amateur telecommunication community.
- Ground station (UHF, VHF or S-band, X-band) is required for satellite operation, which you can build or can even rent/borrow from companies or space agency, etc.
  - Yagi-antenna (cheap but low gain), parabolic antenna, etc.
  - Modulator/demodulator, low noise amplifier, etc.
- Command/telemetry should be defined by satellite developers and software should be coded on ground station’s computer
  - Can expect 6～8 times of 5～12 minutes operations per day
  - Should make operation plans (check items, commands, etc.)

8. Summary

Keep these in mind !

- Create interesting yet achievable mission considering your skill level, satellite size and available resources
- Conduct system level design to define subsystem specifications with weight, size and power budgeting
- Teaming should be well designed considering expertise of your team members
- Follow “water flow” type project management with review meetings and various ground tests, in order to assure proper functionality of your satellite in space
- Make appropriate design of your satellite’s interface with the deployment system (for ISS case, J-SSOD)
- Preparations such as finding a ground station and frequency permission are required in parallel with satellite development